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LAW OFFICES

SUGHRUE, MION, ZINN, MACPEAK & SEAS, PLLC

2100 PENNSYLVANIA AVENUE, N.W.
WASHINGTON, DC 20037-3213
TELEPHONE (202) 293-7060
FACSIMILE (202) 293-7860
www.sughrue.com

March 24, 2000

BOX PATENT APPLICATION

Assistant Commissioner for Patents
Washington, D.C. 20231

Re: Application of Christian DUJARRIC
ROCKET ENGINE NOZZLE COMPRISING A JET SEPARATION CONTROL
SYSTEM
Our Ref. Q58472

Dear Sir:

Attached hereto is the application identified above including 16 sheets of the specification, including claims and abstract, 1 sheet of formal drawings, an executed Assignment and PTO 1595 form, and an executed Declaration and Power of Attorney. Also enclosed is an Information Disclosure Statement and PTO 1449 Form.

Priority is claimed from March 25, 1999 based on French Application No. 9903742. The priority document is enclosed herewith.

A Preliminary Amendment is also enclosed.

After consideration of the Preliminary Amendment, the Government filing fee is calculated as follows:

Total claims	13	-	20	=	x	\$18.00	=	\$0.00
Independent claims	1	-	3	=	x	\$78.00	=	\$0.00
Base Fee								\$690.00

TOTAL FILING FEE	\$690.00
Recordation of Assignment	\$40.00
TOTAL FEE	\$730.00

Checks for the statutory filing fee of \$690.00 and Assignment recordation fee of \$40.00 are attached. You are also directed and authorized to charge or credit any difference or overpayment to Deposit Account No. 19-4880. The Commissioner is hereby authorized to charge any fees under 37 C.F.R. §§ 1.16 and 1.17 and any petitions for extension of time under 37 C.F.R. § 1.136 which may be required during the entire pendency of the application to Deposit Account No. 19-4880. A **duplicate** copy of this transmittal letter is attached.

Respectfully submitted,
**SUGHRUE, MION, ZINN,
MACPEAK & SEAS, PLLC**
Attorneys for Applicant

By: *Robert J. Seas, Jr.*
Robert J. Seas, Jr., Reg. No. 21,092
for
Neil B. Siegel, Reg. No. 25,200

PATENT APPLICATION
Attorney Docket No. Q58472

IN THE UNITED STATES PATENT AND TRADEMARK OFFICE

In re application of

Christian DUJARRIC

Filed: March 24, 2000

For: ROCKET ENGINE NOZZLE COMPRISING A JET SEPARATION CONTROL SYSTEM

PRELIMINARY AMENDMENT

Assistant Commissioner for Patents
Washington, D.C. 20231

Sir:

Prior to examination, please amend the above-identified application as follows:

IN THE CLAIMS:

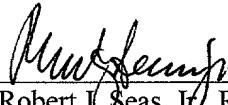
Please amend the claims as follows:

Claim 6,	lines 1-2,	delete "one of the preceding claims" and insert --claim 1--.
Claim 8,	lines 1-2,	delete "one of the preceding claims" and insert --claim 1--.
Claim 10,	lines 1-2,	delete "either of claims 8 and 9" and insert --claim 8--.
Claim 11,	line 1,	delete "one of claims 8 to 10" and insert --claim 8--.

REMARKS

The foregoing amendments are made in order to remove multiple dependencies and avoid the Government surcharge. Entry and consideration of this Amendment is respectfully requested.

Respectfully submitted,



Robert J. Seas, Jr., Reg. No. 21,092
for
Neil B. Siegel, Reg. No. 25,200

SUGHRUE, MION, ZINN,
MACPEAK & SEAS, PLLC
2100 Pennsylvania Avenue, N.W.
Washington, D.C. 20037-3213
Telephone: (202) 293-7060
Facsimile: (202) 293-7860

Date: March 24, 2000

ROCKET ENGINE NOZZLE COMPRISING A JET SEPARATION CONTROL SYSTEM

The subject of the present invention is a
5 rocket engine nozzle, exhibiting a jet separation control system, for example a device for injecting fluid through a wall of the nozzle, so as to induce jet separation in the gases ejected by the nozzle.

An important point in the design of a launcher
10 is the optimization of the performance of its engines. In particular, the nozzle must be designed so as to yield a maximum thrust coefficient compatible with the limits imposed by the other constraints.

The thrust coefficient C of a nozzle is an
15 increasing function of the ratio of the exit area A_e of the nozzle to the area A_t of the throat of the nozzle.

For an upper stage, which is ignited outside the atmosphere, the static pressure of the jet at the exit of the nozzle may be very low. The area ratio
20 $R = A_e/A_t$ of the nozzle is in this case essentially limited by the space available.

On the other hand, when the nozzle operates within the atmosphere, the gases which exit the nozzle cannot expand to below a limit pressure P_{sep} , at which
25 a separation of flow in the nozzle occurs spontaneously.

This jet separation is naturally unstable and generates considerable aerodynamic forces at the moment of ignition and during the initial atmospheric flight,
30 which may even lead to the destruction of the nozzle if the jet separation is too considerable.

As far as engines which are designed to operate right from the ground and to accomplish the major part of their mission outside the atmosphere are concerned,
35 the determination of the ratio A_e/A_t represents a difficult compromise.

Numerous devices have been proposed for controlling jet separation in nozzles.

An up-to-date review of this topic has appeared in particular in the article entitled Advanced Rocket Nozzles by Gerald Hagemann et al., published in the Journal of Propulsion and Power, vol. 14 No. 5, 5 September-October 1998, pages 620 to 634.

This deals in particular with "dual-bell" nozzles, nozzles with fixed or temporary inserts, two-position or extendible nozzles, external expansion nozzles, so-called expansion/deflection nozzles, 10 nozzles exhibiting a variable throat area, and finally dual-mode nozzles.

The control of jet separation in a nozzle with the aid of secondary injection of gas has also been proposed, but this secondary injection has the effect 15 of preserving axial symmetry of the flow. This technique is recalled in point 4, page 626 of the aforesaid article.

Experiments carried out on an RL10 engine and implementing passive injection are described in the 20 article entitled "Altitude Compensating Nozzle Evaluation" by R.C. PARSLEY et al., published in the proceedings of the 28th Joint Propulsion Conference and Exhibit, 6 to 8 July 1992, Nashville, Tennessee, pages 1 to 6.

25 Finally, American patent US 3 925 982 (Martin Marietta Corporation) describes a rocket engine exhibiting a high nozzle area ratio and which is equipped with a device for active secondary injection exhibiting a shock generating ring which is intended to 30 control jet separation, by forcing the boundary layer of the primary gas jet to separate uniformly from the wall of the nozzle.

This is achieved with the aid of a large number 35 of injection points which are distributed around the circumference of the nozzle. These injection points are closely spaced, and they inject a secondary gas jet radially and inwardly of the nozzle so as to effect jet separation which is invariant with any rotation about the axis of the nozzle.

Alternatively, this jet separation can be achieved via a continuous slot extending over the entire circumference of the nozzle.

5 The theory of jet separation has been recalled in the recent article by G.L. ROMINE entitled "Nozzle Flow Separation" published in the AIAA Journal, vol. 36, No. 9, Sept. 1998, pp. 1618-1625.

10 The theory of secondary injection has been set out in the article entitled "Some aspects of gaseous secondary injection with application to thrust vector control" by R.D. GUHSE et al., published in proceedings No. 71-750 of the AIAA/SAE 7th Propulsion Joint Specialist Conference of Salt Lake City, 14-18 June 1971, pages 1 to 8.

15 The known techniques of secondary injection, which involve jet separation exhibiting axial symmetry, that is to say which is invariant about any rotation about the axis of the nozzle, exhibit the following problems:

20 - active secondary injection is difficult to implement, given that the mass flux which is required for effective generation of axial symmetric jet separation is high;

25 - passive secondary injection which implements ventilation of the nozzle is operational only within a limited range of differential pressure, which implies that in order to obtain a nozzle which operates at all altitudes, its porosity must be continuously variable as a function of external pressure and of the 30 operational parameters of the engine, this being hardly compatible with the nozzle construction constraints.

35 One of the drawbacks of secondary injections with axial symmetry, such as for example that described in the aforesaid American patent, is that under certain engine operating conditions, the jet separation commences at a random point on the injection ring, and whose position, which depends on the upstream disturbances, is unstable.

An object of the invention is to propose a separation control system, especially through secondary injection which avoids such instability.

Another object of the invention is to reduce 5 the unsteady loads applied to nozzles and hence to reduce the necessary mechanical strength of the nozzles and the engine mountings, thereby making it possible to reduce their mass.

Another object of the invention is to allow the 10 installation on rocket engines used right from the ground, of nozzles with a higher expansion ratio and hence to enable an overall improvement in the performance of these engines.

Another object of the invention is to minimize 15 the total secondary injection flux required to obtain stable separation.

Another object of the invention is to control the separation of the jet during ignition of the engine on the ground.

20 Another object of the invention is to be able to facilitate the control of jet separation as a function of altitude.

At least one of the aforesaid objects of the present invention is achieved through a rocket engine 25 nozzle comprising a system for controlling jet separation, wherein said control system exhibits a plurality of separation triggering elements arranged in such a way as to generate, from mutually spaced initiation points, distinct zones of jet separation, so 30 as to form a three-dimensional separation of the flow.

According to a first variant, the invention relates to an ejection nozzle for a rocket engine exhibiting a device for injecting fluid through a wall 35 of the nozzle, so as to induce jet separation in the gases ejected by the nozzle, in which the control system is a fluid-injection device which exhibits in at least one injection cross section perpendicular to the axis of the nozzle, at least two independent injection orifices distributed over the perimeter of the wall of

the nozzle, each injection orifice constituting a said separation triggering element inducing a said distinct zone of jet separation.

According to the invention, each separation 5 triggering element, for example each injection orifice initiates and maintains locally the separation of the jet, thereby remedying the aforesaid instability. This arrangement is essentially different from that which is described in the aforesaid American patent, for which 10 the injection orifices are tightly distributed over the perimeter of the nozzle so as to generate a flow separation which is invariant with any rotation of the axis of the nozzle, and which operates in a manner equivalent to that of a ring exhibiting a continuous 15 slot extending over the entire circumference of the nozzle.

It is advantageous for the injection orifices, for example two in number or else three in number, to be uniformly distributed over the perimeter of the wall 20 of the nozzle. This makes it possible to avoid to a large extent the occurrence of transverse forces applied to the nozzle.

Said injection cross section is advantageously arranged at a distance D from the throat of the nozzle 25 which is substantially less than the distance D_0 of spontaneous separation of the jet.

Said nozzle cross section is preferably chosen at a level where the static pressure P of the jet is substantially greater than the natural separation 30 pressure of the jet P_{sep} , for example $P = 2 P_{sep}$.

An injection device can exhibit a plurality of injection cross sections situated at different distances from the throat, and a distributing device for feeding one or other of the injection cross 35 sections, in such a way as to take into account, in a manner known per se, the variation as a function of altitude of the cross section where a so-called spontaneous separation of the jet occurs.

According to an embodiment making it possible to control jet separation during ignition of the engine on the ground, the flow control system exhibits an external stabilizing device integral with a ground-based installation and which exhibits, on the one hand, a number $N (N \geq 2)$ of injection tubes (for example parallel to the axis of the nozzle) each of which constitutes a said separation triggering element, and which are distributed, preferably downstream of the nozzle, in such a way as to direct in counter-current to the main stream of the nozzle stabilizing fluid jets toward N impact points situated downstream of the throat of the nozzle, and on the other hand, a device for feeding the injection tubes so as to feed them with fluid for a predetermined transient duration of ignition before takeoff, with a flow rate which is sufficient for each impact point to induce a different zone of jet separation of the nozzle.

The injection tubes are preferably arranged at the outlet of the nozzle exit.

The injection points of the injector of the external stabilizing device are preferably uniformly distributed over the perimeter of the wall of the nozzle. They are advantageously two in number (diametrically opposed) or three in number (distributed at around 120° over the perimeter of the nozzle).

Other characteristics and advantages of the invention will become more apparent on reading the description which will follow, given by way of non-limiting example in conjunction with the drawings herein appended, in which:

- Figure 1 represents a device for implementing the present invention; and
- Figure 2 represents an additional device according to the present invention which is implemented during engine start-up.

As shown in Figure 1, a nozzle, designated by the general label 1, exhibits a combustion chamber 2, a

throat 3, and a divergent nozzle body 4 which terminates in an exit cross section 8.

Over the perimeter of the divergent portion 4 of the nozzle, and in a cross section 7 situated in a 5 plane, perpendicular to the axis of the nozzle, where the static pressure P of the jet is substantially greater than the nozzle separation pressure P_{sep} , are arranged injection orifices 5 able to direct radially inward a jet of a fluid, for example the combustion 10 gases originating from the turbopumps of the engine.

The flow separation which is generated by these orifices 5, does not exhibit axial symmetry, but on the contrary it is three-dimensional. This is because each 15 of the injection points 5, represented here as three in number and distributed uniformly at 120° around the contour of the body 4 of the nozzle, induces a region of separation 6 of the stream exiting the nozzle. Owing to the determination of a limited number of injection points 5 which induce an equal number of separation 20 regions 6, the position of the points of initial separation is not indeterminate and this makes it possible to solve the problem of instability.

Furthermore, by reason of the uniform distribution of the injection points 5 around the 25 circumference of the nozzle body 4 in the plane 7, the resultant of the lateral forces which are exerted on the nozzle and which, in the prior art is unstable, remains close to the axis of the nozzle.

The cross section in which the injection is 30 carried out is chosen in such a way as to be of slightly smaller area than that of the cross section at which spontaneous stream separation would be apt to occur at low altitude.

The device described makes it possible to 35 obtain three-dimensional separation exhibiting a plurality of separated regions 6 which may possibly remerge downstream toward the exit 8 of the nozzle.

In theory, the number of injection points 5 could be equal to just 2 so as to make it possible to

5 maintain a symmetric thrust for the nozzle. The number of three injection points 5 seems however to be a preferable choice in order to avoid accidental separation of half the nozzle which could arise on ignition.

10 Moreover, a higher number than three injection points may be envisaged, but this does not afford any appreciable advantage. In any event, the number and the spacing of the injection points must be chosen in such a way as to avoid any continuity of jet separation, which would amount in fact to the operating conditions equivalent to that of a uniform ring (see the aforesaid American patent US 3 925 982).

15 The manner of operation of the invention can be explained as follows:

20 The gases which are generated in the high-pressure combustion chamber 2 are, after passing through the throat 3, subjected to an expansion in the ejection nozzle 4 and the static pressure decreases at the same time as the jet is directed toward the exit 8 of the nozzle.

25 When a rocket engine operates at the static pressure which prevails at sea level, and without implementing secondary injection, as soon as the static pressure of the jet approaches the normal pressure for separation, the jet in the nozzle is apt to separate spontaneously, but such separation is unstable and occurs in a random place of a cross section of the body of the nozzle 4, possibly creating appreciable unsteady 30 forces.

35 The injection orifices 5 which direct a secondary stream radially inward according to the invention create an obstacle to the main jet locally, thereby creating an arc-shaped shock wave in respect of the incident supersonic jet. This arc-shaped shock zone interacts with the boundary layer, in which it creates an increase in pressure just upstream of the injection point 5, thereby inducing local separation of the boundary layer at the points 9. Given that the boundary

layer was already under conditions under which it was near to spontaneous separation, the jet of the nozzle cannot reattach itself to the wall of the body 4 and the separation of the boundary layer spreads so as to 5 adopt for each injection point 5 a conical configuration as shown by the dashed drawing of the separation regions 6 in Figure 1. The vertex of the cones 6 is constituted by the points 9 of initiation of jet separation. The three injection points 5 create, 10 starting from the initiation points 9, three substantially identical cones 6 which are apt to remerge downstream so as to form an entirely separated jet at the exit 8 of the nozzle.

Given that the points of initiation 9 where the 15 jet separations occur are imposed geometrically by the position of the three injection points 5, the symmetry of revolution is broken and the points 9 of initiation of the separation are stable over time. The shocks which are created due to the separation of the boundary 20 layer relative to the wall of the body 4 also remain localized and the residual vibrations due to these shocks are of low amplitude, as are the residual unsteady forces.

Furthermore, the injected flux which is 25 required for achieving separation according to the invention is in principle smaller than in the case of secondary injection with axial symmetry of the prior art. This is because such an injection of the prior art implements a large number of orifices along a circular 30 cross section of the nozzle. The separation of the jet requires locally at each of these numerous orifices the same minimum flow rate as each of the few orifices used according to the present invention. In the present invention, separation, which is achieved only onward of 35 the localized points of initiation 9, tends thereafter to self-propagate along the periphery of the nozzle along with the widening and merging of the cones 6.

The process according to the invention is particularly beneficial in respect of engines

exhibiting a gas generator. Such an engine exhibits turbopumps which discharge hot gases at a pressure greater than atmospheric pressure. The Vulcain engine is of this type. The gases from the turbine of the 5 Vulcain 1 engine are expelled from the engine. In the Vulcain 2 engine, pipelines are already installed for reinjecting the turbine exit gases into the divergent portion 4 with a uniform distribution along a cross section of the nozzle, so as to cool the divergent 10 portion, but nevertheless without achieving control of jet separation.

These gases can be reinjected at a few points only into the divergent body 4 of the nozzle to achieve jet separation according to the invention.

15 The present invention can be adapted to the Vulcain 2 engine with minimal modifications. It is sufficient to modify the injection ring so that it exhibits for example three injection points 5 instead of a uniform distribution of injection. Furthermore, 20 the divergent portion 4 of the nozzle, which currently exhibits for the Vulcain 2 engine an area ratio R equal to 60 for a specific impulse of 433 seconds, could be replaced with a divergent portion exhibiting an area ratio R of the order of 140. The film-based cooling 25 function could be replaced with radiative cooling, by virtue of a carbon/carbon nozzle extension known per se.

For the Vulcain 2 engine, the value of P_{sep} is of the order of 0.22 bar and the recommended location 30 for siting the points of injection 5 is the cross section for which the pressure is equal to 0.4 bar. This corresponds to a Mach number of 4 and an area ratio R of around 26. The location is not very different from the current location of the injection 35 ring. The anticipated increase in the specific impulse is of the order of 12 seconds.

With altitude, the location of the source cross section where natural jet separation occurs migrates

progressively downstream as the external pressure described decreases.

Furthermore, outside the atmosphere, secondary injection exhibits no benefit, and even penalizes the 5 performance of the engine.

A minimal procedure is to deactivate secondary injection when the rocket leaves the atmosphere. In the Vulcain 2 engine, it is sufficient to add a valve which switches from a local injection mode to a distributed 10 mode (film-based cooling).

Another solution is to arrange different injection points 5 which are activated in succession in such a way as to optimize the operation of the nozzle at each instant. A solution of this type has already 15 been proposed, but for injection at continuous rings by the aforesaid patent US 3 925 982.

The ignition of the engine constitutes a tricky phase during which pressure transients are apt to exist. It is therefore desirable to minimize the 20 considerable asymmetries of pressure which are apt to occur and which are apt to engender a high level of stress in the nozzle. The theoretically least favorable case is that for which the jet of the nozzle is momentarily entirely attached to the wall over one half 25 of the nozzle and entirely separated over the other half thereof.

Owing to the violence of the unsteady fluctuations in the flows during start-up of a rocket engine, it is preferable, during this start-up, to use 30 another embodiment which will be described hereinbelow.

Indeed, it is possible to solve the problem with injection of a control fluid during the very short start-up time of the engine, which is of the order of a second. The point of impact of the injected fluid is 35 close to the throat 3 of the nozzle, for example a distance of the order of 0.1 D1 from the throat 3, D1 designating the length of the body of the nozzle 4, given that the pressure of the chamber is lower than when the engine is at full throttle.

Finally, given that the jet is not organized, the stabilizing of the system requires a massive injection of fluid.

The device for stabilization on blast-off is represented in Figure 2. It can be used independently or otherwise of the jet separation device. It implements a plurality of injection tubes 10 parallel or otherwise to the axis of the nozzle and arranged downstream of the nozzle exit 8 and directed toward impact points 12. These tubes 10 propagate fluid jets 11 in counter-current to the main stream, the points of impact 12 of these jets being situated slightly downstream of the throat 3 of the nozzle, for example a distance from the throat 3 equal to 0.1 D1. These points of impact 12 distributed uniformly at one and the same distance from the throat 3 of the nozzle produce a similar effect to that of the injection points 5, with the difference however that the fluid, for example liquid nitrogen, which is projected creates a separation at each point of impact 12 through a mass entrainment effect along the boundary layer. The points of separation of the hot gases of the jet from the nozzle are stable by reason of the existence of the impact points 12. It is advantageous to effect this injection with liquid nitrogen, since the counter-current injection rate may be very high (for example 30 kg/s for each injection point for the Vulcain 2 engine) during the short instant which proves to be necessary. Furthermore, the liquid nitrogen is transformed into gas when it encounters the hot gases originating from the combustion chamber 12, which means that the mass flux thus added artificially helps to reduce the phenomenon of spontaneous separation. Once full thrust has been established, the nitrogen jet no longer penetrates into the body of the nozzle and it no longer has an influence on the operation of the engine. The stabilizing device is a ground-based device which is generally arranged downstream of the exit 8 of the nozzle and which requires no modification of the engine

or of the launcher. It is apt to be used with nozzles which do or do not exhibit an injection device as represented in Figure 1.

The invention is not limited to the exemplary 5 embodiments described. In particular, jet separation could be initiated for example onward of a plurality of retractable inserts made of refractory material introduced radially into the wall of the nozzle.

CLAIMS

1. A rocket engine nozzle comprising a system for controlling jet separation of the flow in the nozzle, wherein said control system exhibits a plurality of separation triggering elements (5, 10) arranged in such a way as to generate, from mutually spaced initiation points (9), distinct zones (6) of jet separation, so as to form a three-dimensional separation of the flow.

2. The nozzle as claimed in claim 1, wherein the flow control system exhibits a device for injecting fluid through a wall of the nozzle, which exhibits, in at least one injection cross section substantially perpendicular to the axis of the nozzle, at least two independent injection orifices (5) distributed over the perimeter of the wall of the nozzle, each injection orifice (5) constituting a said separation triggering element inducing a said distinct zone (6) of jet separation.

3. The nozzle as claimed in claim 2, wherein the injection orifices (5) are uniformly distributed over the perimeter of the wall of the nozzle (4).

4. The nozzle as claimed in claim 3, wherein the injection orifices (5) are two in number and are diametrically opposed.

5. The nozzle as claimed in claim 3, wherein the injection orifices (5) are 3 in number and are arranged at substantially 120° to one another over the perimeter of the nozzle (4).

6. The nozzle as claimed in one of the preceding claims, wherein said injection cross section is arranged at a distance D from the throat (3) of the nozzle which is substantially less than the distance D_0 of spontaneous separation of the flow.

7. The nozzle as claimed in claim 6, wherein the injection device exhibits a plurality of injectors (5) situated at different distances D, and a distributing device for feeding one or other of said injection cross sections (5), in such a way as to take into account the

variation of said distance D_0 as a function of altitude.

8. The nozzle as claimed in one of the preceding claims, wherein the flow control system exhibits an external stabilizing device integral with a ground-based installation and which exhibits, on the one hand, a number $N (N \geq 2)$ of injection tubes (10) each of which constitutes a said separation triggering element, and which are distributed, preferably downstream of the nozzle (4), in such a way as to direct in counter-current to the main stream of the nozzle a stabilizing fluidic stream toward N impact points (12) situated downstream of the throat (3) of the nozzle (4), and on the other hand, a device (AL) for feeding the injection tubes (10) so as to feed them with fluid for a predetermined transient duration of ignition before takeoff, with a flow rate which is sufficient for each impact point (12) to induce a different zone of jet separation of the nozzle.

9. The nozzle as claimed in claim 8, wherein the injection tubes (10) are parallel to the axis of the nozzle.

10. The nozzle as claimed in either of claims 8 and 9, wherein the injection tubes (10) are arranged at the outlet of the nozzle (4) exit (8).

11. The nozzle as claimed in one of claims 8 to 10, wherein the impact points (12) of the external stabilizing device are uniformly distributed over the perimeter of the wall of the nozzle.

12. The nozzle as claimed in claim 11, wherein the impact points (12) of the external stabilizing device are two in number and are diametrically opposed.

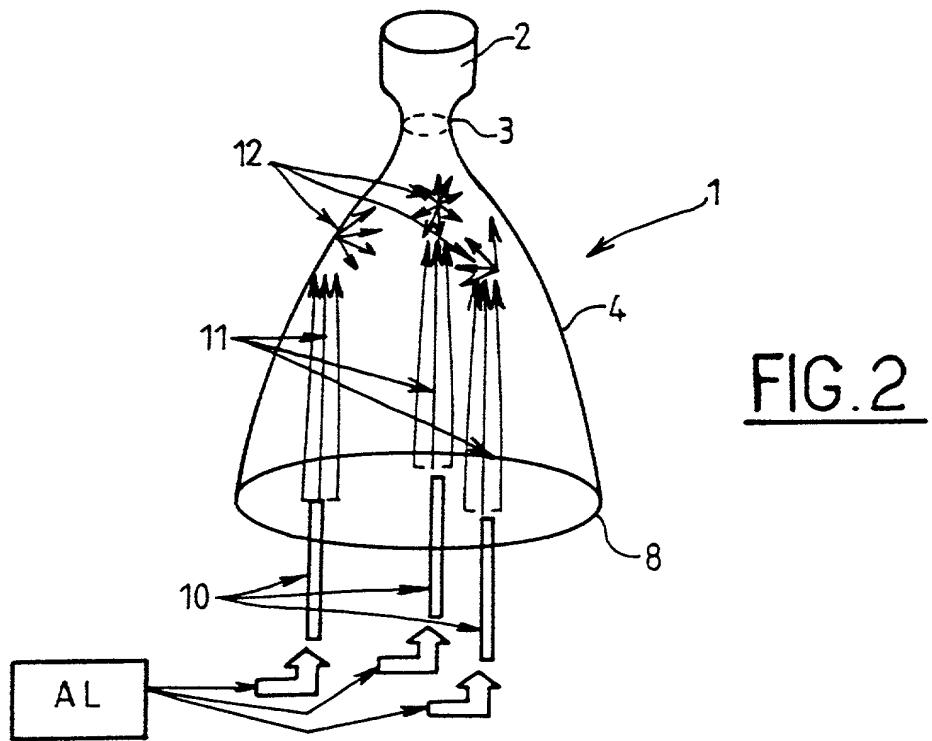
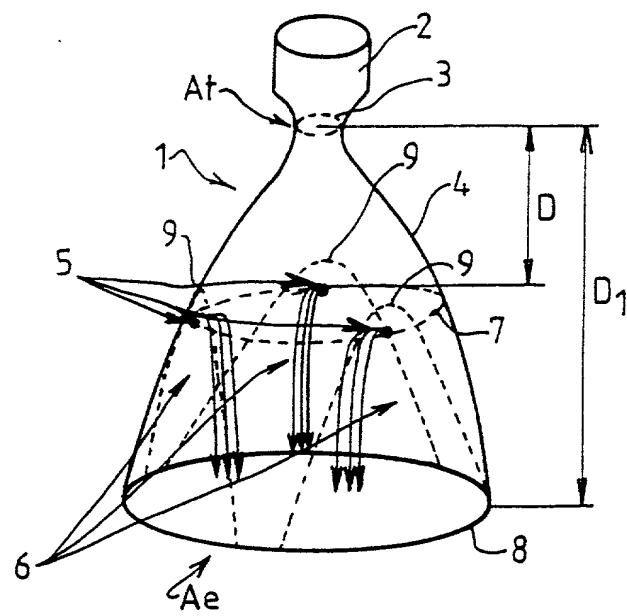
13. The nozzle as claimed in claim 11, wherein the impact points (12) of the external device are three in number and are arranged at substantially 120° to one another over the perimeter of the nozzle.

ABSTRACT

The invention relates to a rocket engine nozzle comprising a system for controlling jet separation of the flow in the nozzle, wherein said control system exhibits a plurality of separation triggering elements (5, 10) arranged in such a way as to generate, from mutually spaced initiation points (9), distinct zones (6) of jet separation, so as to form a three-dimensional separation of the flow.

The flow control system can exhibit at least two triggering elements (5, 10).

Figure 1



DECLARATION AND POWER OF ATTORNEY

As a below named inventor, I hereby declare that my residence, post office address and citizenship are as stated below next to my name: that I verily believe I am the original, first and sole inventor (if only one name is listed below) or an original, first and joint inventor (if plural names are listed below) of the subject matter claimed and for which a patent is sought in the application entitled:

ROCKET ENGINE NOZZLE COMPRISING A JET SEPARATION CONTROL SYSTEM

which application is:

the attached application
(for original application)

application Serial No. _____
filed _____, and amended on _____

(for declaration not accompanying application)

that I have reviewed and understand the contents of the specification of the above-identified application, including the claims, as amended by any amendment referred to above; that I acknowledge my duty to disclose information of which I am aware which is material to the patentability of this application under 37 C.F.R. 1.56, that I hereby claim priority benefits under Title 35, United States Code §119, §172 or §365 of any provisional application or foreign application(s) for patent or inventor's certificate listed below and have also identified on said list any foreign application for patent or inventor's certificate on this invention having a filing date before that of any foreign application on which priority is claimed:

Application Number	Country	Filing Date	Priority Claimed (yes/no)
99 03742	FRANCE	25, MARCH 1999	YES

I hereby claim the benefit of Title 35, United States Code §120 of any United States application(s) listed below and, insofar as the subject matter of each of the claims of this application is not disclosed in a listed prior United States application in the manner provided by the first paragraph of Title 35, United States Code, §112, I acknowledge my duty to disclose any information material to the patentability of this application under 37 C.F.R. 1.56 which occurred between the filing date of the prior application and the national or PCT international filing date of this application:

Application Serial No.	Filing Date	Status (patented, pending, abandoned)
------------------------	-------------	--

I hereby appoint John H. Mion, Reg. No. 18,879; Thomas J. Macpeak, Reg. No. 19,292; Robert J. Seas, Jr., Reg. No. 21,092; Darryl Mexic, Reg. No. 23,063; Robert V. Sloan, Reg. No. 22,775; Peter D. Olexy, Reg. No. 24,513; J. Frank Osha, Reg. No. 24,625; Waddell A. Biggart, Reg. No. 24,861; Louis Gubinsky, Reg. No. 24,835; Neil B. Siegel, Reg. No. 25,200; David J. Cushing, Reg. No. 28,703; John R. Inge, Reg. No. 26,916; Joseph J. Ruch, Jr., Reg. No. 26,577; Sheldon I. Landsman, Reg. No. 25,430; Richard C. Turner, Reg. No. 29,710; Howard L. Bernstein, Reg. No. 25,665; Alan J. Kasper, Reg. No. 25,426; Kenneth J. Burchfiel, Reg. No. 31,333; Gordon Kit, Reg. No. 30,764; Susan J. Mack, Reg. No. 30,951; Frank L. Bernstein, Reg. No. 31,484; Mark Boland, Reg. No. 32,197; William H. Mandir, Reg. No. 32,156; Scott M. Daniels, Reg. No. 32,562; Brian W. Hannon, Reg. No. 32,778; Abraham J. Rosner, Reg. No. 33,276; Bruce E. Kramer, Reg. No. 33,725; Paul F. Neils, Reg. No. 33,102; Brett S. Sylvester, Reg. No. 32,765; and Robert M. Masters, Reg. No. 35,603, my attorneys to prosecute this application and to transact all business in the Patent and Trademark Office connected therewith, and request that all correspondence about the application be addressed to **SUGHRUE, MION, ZINN, MACPEAK & SEAS, PLLC**, 2100 Pennsylvania Avenue, N.W., Washington, D.C. 20037-3202.

I hereby declare that all statements made herein of my own knowledge are true and that all statements made on information and belief are believed to be true; and further that these statements were made with the knowledge that willful false statements and the like so made are punishable by fine or imprisonment, or both, under Section 1001 of Title 18 of the United States Code and that such willful false statements may jeopardize the validity of the application or any patent issuing thereon.

Date 9/3/2000 First Inventor Christian Francois Michel DUJARRIC

First Name Middle Initial Last Name

Residence PARIS Signature  Christian F.M. DUJARRIC

Post Office Address: 52, rue Michel Ange, 75016 PARIS, FRANCE

Citizenship FRENCH